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Lateral Control Formulas

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List of 10 Lateral Control Formulas

Lateral Control

1) Aileron Control Effectiveness given Aileron Deflection

$$\text{fx } \tau = \frac{C_l}{C_{l\alpha} \cdot \delta_a}$$

[Open Calculator !\[\]\(a870788d6ed9b8fd294b7654a8c8526b_img.jpg\)](#)

$$\text{ex } 0.663636 = \frac{0.073}{0.02 \cdot 5.5\text{rad}}$$

2) Aileron Deflection given Aileron Lift Coefficient

$$\text{fx } C_l = \frac{2 \cdot C_{l\alpha w} \cdot \tau \cdot \delta_a}{S \cdot b} \cdot \int (c \cdot x, x, y_1, y_2)$$

[Open Calculator !\[\]\(c50c8b7b2cc2cf9ff925edec0ee94c0d_img.jpg\)](#)

$$\text{ex } 0.073097 = \frac{2 \cdot 0.23 \cdot 0.66 \cdot 5.5\text{rad}}{17\text{m}^2 \cdot 200\text{m}} \cdot \int (2.1\text{m} \cdot x, x, 1.5\text{m}, 12\text{m})$$

3) Aileron Section lift Coefficient given Aileron Deflection

$$\text{fx } C_l = C_{l\alpha} \cdot \left(\frac{d\alpha}{d\delta_a} \right) \cdot \delta_a$$

[Open Calculator !\[\]\(f60b7a900783ac3fd531bfd9c111be6d_img.jpg\)](#)

$$\text{ex } 0.073333 = 0.02 \cdot \left(\frac{3.0\text{rad}}{4.5\text{rad}} \right) \cdot 5.5\text{rad}$$



4) Aileron Section Lift Coefficient given Control Effectiveness

$$fx \quad C_l = C_{l_\alpha} \cdot \tau \cdot \delta_a$$

[Open Calculator !\[\]\(cbe80b694ebd74fcfe136a095b608235_img.jpg\)](#)

$$ex \quad 0.0726 = 0.02 \cdot 0.66 \cdot 5.5rad$$

5) Deflection Angle given Lift Coefficient

$$fx \quad \delta_a = \frac{C_l}{C_{l_\alpha} \cdot \tau}$$

[Open Calculator !\[\]\(3e2231b1ad3ca8da8658228c00dd08e0_img.jpg\)](#)

$$ex \quad 5.530303rad = \frac{0.073}{0.02 \cdot 0.66}$$

6) Lift Coefficient Slope Roll Control

$$fx \quad C_{l_\alpha} = \frac{C_l}{\delta_a \cdot \tau}$$

[Open Calculator !\[\]\(0d5ec72f61334709c3fc9450209b754f_img.jpg\)](#)

$$ex \quad 0.02011 = \frac{0.073}{5.5rad \cdot 0.66}$$

7) Lift Coefficient with respect to Roll Rate

$$fx \quad Cl = - \left(\frac{2 \cdot p}{S_r \cdot b \cdot u_0} \right) \cdot \int \left(Cl_\alpha \cdot c \cdot x^2, x, 0, \frac{b}{2} \right)$$

[Open Calculator !\[\]\(b64b40baaee5acddc1eab8538ba84754_img.jpg\)](#)

$$ex \quad 0.038043 = - \left(\frac{2 \cdot 0.5rad/s^2}{184m^2 \cdot 200m \cdot 50m/s} \right) \cdot \int \left(-0.1 \cdot 2.1m \cdot x^2, x, 0, \frac{200m}{2} \right)$$



8) Lift given Roll Rate 

$$\text{fx } L = -2 \cdot \int \left(Cl_{\alpha} \cdot \left(\frac{p \cdot x}{u_0} \right) \cdot Q \cdot c \cdot x, x, 0, \frac{b}{2} \right)$$

Open Calculator 

ex

$$770\text{N} = -2 \cdot \int \left(-0.1 \cdot \left(\frac{0.5\text{rad/s}^2 \cdot x}{50\text{m/s}} \right) \cdot 0.55\text{rad/s}^2 \cdot 2.1\text{m} \cdot x, x, 0, \frac{200\text{m}}{2} \right)$$

9) Roll Control Power 

$$\text{fx } Cl_{\delta\alpha} = \frac{2 \cdot C_{l_{\alpha w}} \cdot \tau}{S \cdot b} \cdot \int (c \cdot x, x, y_1, y_2)$$

Open Calculator 

$$\text{ex } 0.01329\text{rad} = \frac{2 \cdot 0.23 \cdot 0.66}{17\text{m}^2 \cdot 200\text{m}} \cdot \int (2.1\text{m} \cdot x, x, 1.5\text{m}, 12\text{m})$$

10) Roll Damping Coefficient 

$$\text{fx } Cl_p = -\frac{4 \cdot C_{l_{\alpha w}}}{S \cdot b^2} \cdot \int \left(c \cdot x^2, x, 0, \frac{b}{2} \right)$$

Open Calculator 

$$\text{ex } -0.947059 = -\frac{4 \cdot 0.23}{17\text{m}^2 \cdot (200\text{m})^2} \cdot \int \left(2.1\text{m} \cdot x^2, x, 0, \frac{200\text{m}}{2} \right)$$









Variables Used

- **b** Wingspan (Meter)
- **c** Chord (Meter)
- **C_l** Lift Coefficient Roll Control
- **C_{l α}** Lift Coefficient Slope Roll Control
- **C_{l α w}** Derivative of Wing Lift Coefficient
- **Cl** Lift Coefficient with respect to Roll Rate
- **Cl_p** Roll Damping Coefficient
- **Cl _{α}** Lift Curve Slope
- **Cl _{$\delta\alpha$}** Roll Control Power (Radian)
- **d α** Rate of change of Angle of Attack (Radian)
- **d δ_a** Rate of change of Deflection of Aileron (Radian)
- **L** Lift with respect to Roll Rate (Newton)
- **p** Roll Rate (Radian per Square Second)
- **Q** Pitch Rate (Radian per Square Second)
- **S** Wing Area (Square Meter)
- **S_r** Wing reference Area (Square Meter)
- **u₀** Reference Velocity across X Axis (Meter per Second)
- **y₁** Initial Length (Meter)
- **y₂** Final Length (Meter)
- **δ_a** Deflection of Aileron (Radian)
- **T** Flap Effectiveness Parameter



Constants, Functions, Measurements used

- **Function:** **int**, $\text{int}(\text{expr}, \text{arg}, \text{from}, \text{to})$
The definite integral can be used to calculate net signed area, which is the area above the x -axis minus the area below the x -axis.
- **Measurement:** **Length** in Meter (m)
Length Unit Conversion 
- **Measurement:** **Area** in Square Meter (m^2)
Area Unit Conversion 
- **Measurement:** **Speed** in Meter per Second (m/s)
Speed Unit Conversion 
- **Measurement:** **Force** in Newton (N)
Force Unit Conversion 
- **Measurement:** **Angle** in Radian (rad)
Angle Unit Conversion 
- **Measurement:** **Angular Acceleration** in Radian per Square Second (rad/s^2)
Angular Acceleration Unit Conversion 



Check other formula lists

- [Directional Stability Formulas](#) 
- [Lateral Control Formulas](#) 

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